

ABSTRACT OF THE DISCLOSURE

In the blade structure in a gas turbine, front-edge including angles are made large. As a result, a curve of a relative relationship between incidence angles i_{c1} and 5 i_{s1} and pressure loss becomes mild. Entrance metal angles are made small. As a result, it becomes possible to make the incidence angles small. Chord length of a chip portion of a moving blade is made large. As a result, it becomes possible to make small the deceleration on a rear surface 10 of the chip portion of the moving blade. Accordingly, it becomes possible to make the pressure loss small, and therefore, it becomes possible to improve the turbine efficiency.